

Experimental and numerical study on the compressive failure of multidirectional laminates made of Carbon Fibre Reinforced Polymers

F. Laurin, C. Fougrouse, A. Mavel

Modelling of longitudinal compression failure of UD ply

Physical mechanisms ?

Fibre kinking



Experimental evidences

Instability of carbon fibres in polymer matrix observed at microscopic scale (*structural effect*)

[Argon 72]
[Vogler 01]
[Gutkin 10] ...

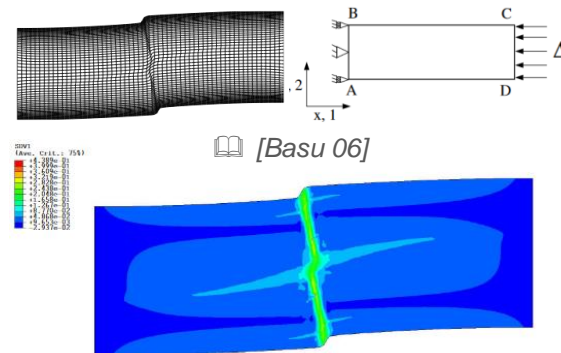
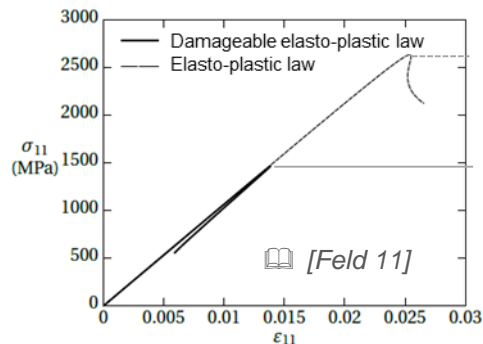
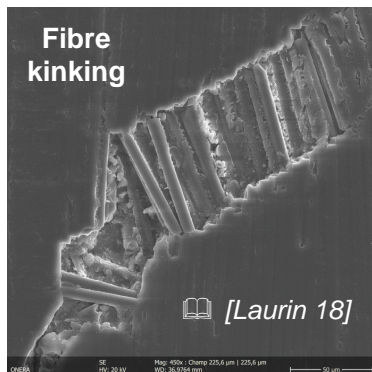


Micromechanical modelling

Models at microscale depend on:

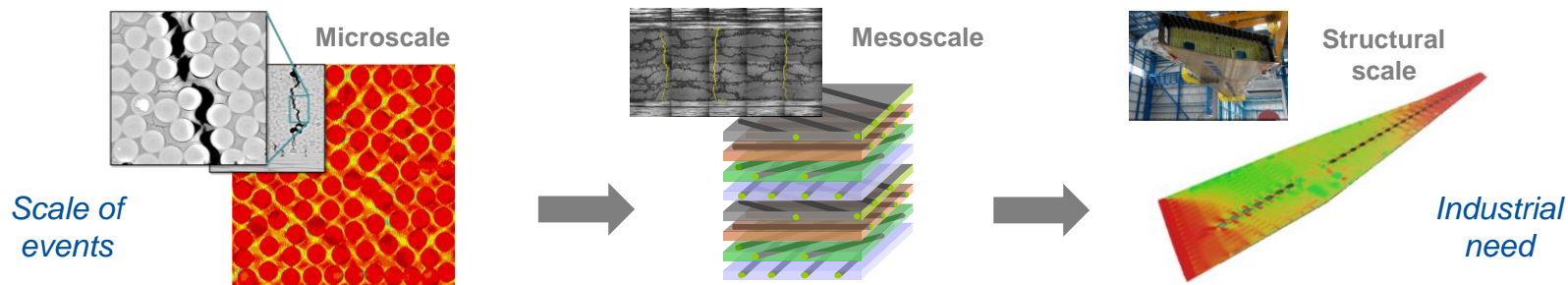
- Mechanical properties of fibres **and** matrix (*and behaviour*)
- Applied loadings (*compression, bending ...*)
- Boundary conditions (*inner or outer plies*)

[Lee 99]
[Drapier 99]
[Pimenta 09]
[Feld 11]...

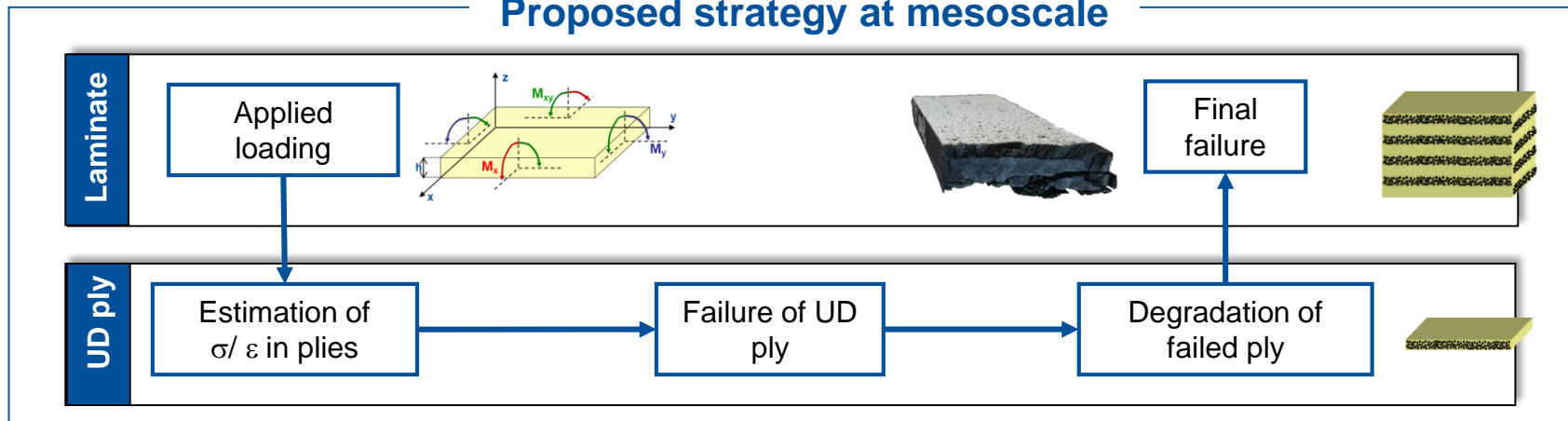


Modelling of longitudinal compression failure of UD ply

What is the relevant modelling scale?

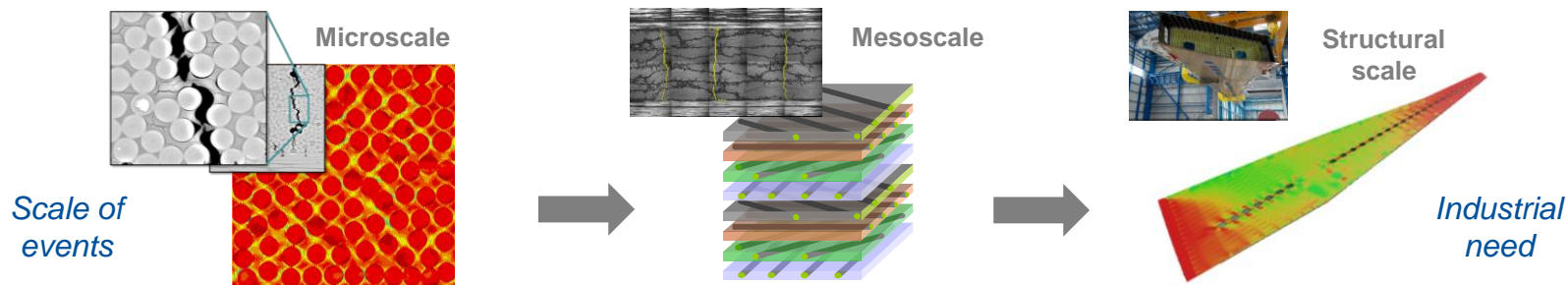


Proposed strategy at mesoscale

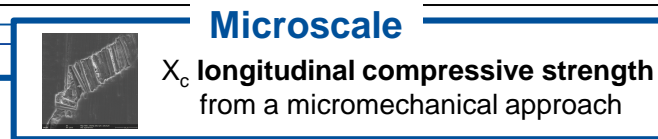
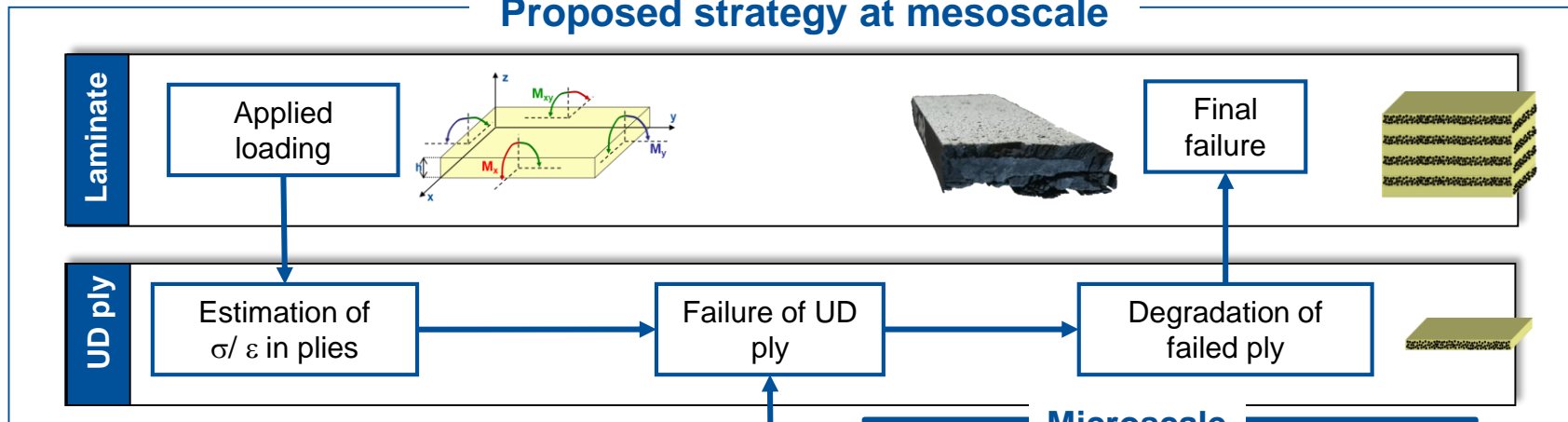


Modelling of longitudinal compression failure of UD ply

What is the relevant modelling scale?



Proposed strategy at mesoscale



Mesoscale failure approach

Estimation of macroscopic failure

Onera Failure Model



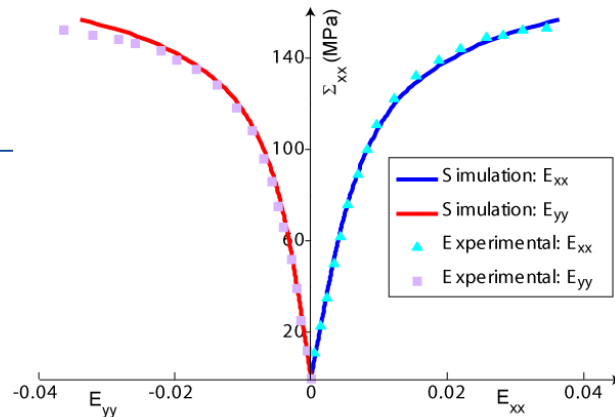
Key points

- Thermal residual stresses [Huchette 05]
- **Non-linear longitudinal elastic behaviour** [Laurin 07]
- **Viscoelastic behaviour** [Schieffer 03, Laurin 05]
- Influence of ply thickness on IFF strengths [Parvizi 78, Chang 87]
- Advanced failure criteria [Puck 02, Laurin 07, Pinho 13, Carrere 13]



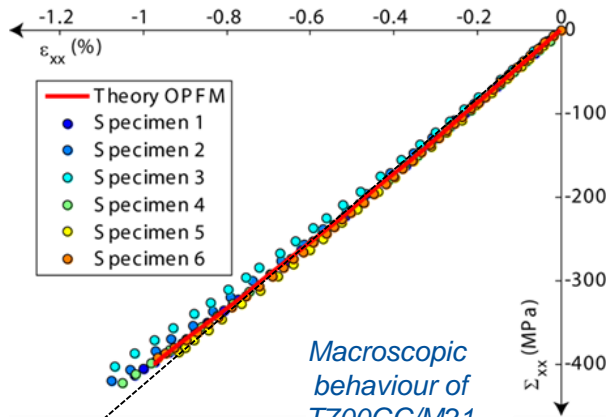
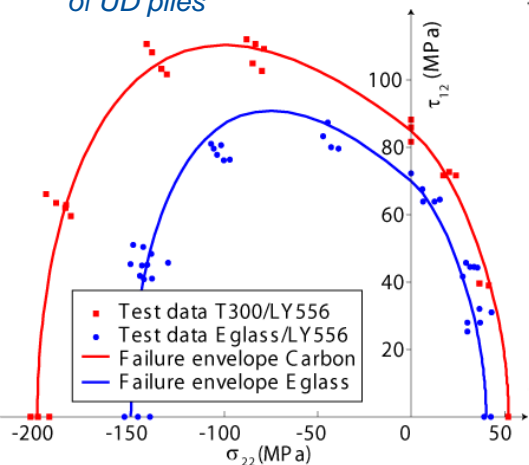
Identification and validation

- Already identified and validated on a **Carbon/Epoxy T700GC/M21** [Huchette 05, Laurin 07]



Macroscopic behaviour of T700GC/M21 [±45°]_{2s}

Failure envelopes of UD plies



Macroscopic behaviour of T700GC/M21 [45/90/-45/0]_s

Mesoscale failure approach

Estimation of macroscopic failure

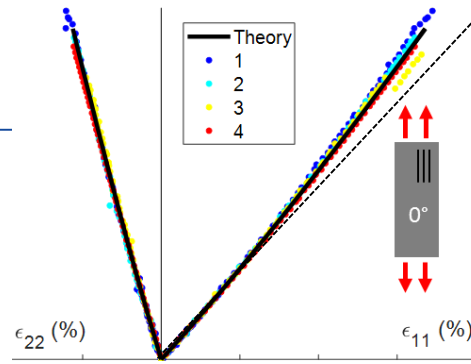
Onera Failure Model

Key points

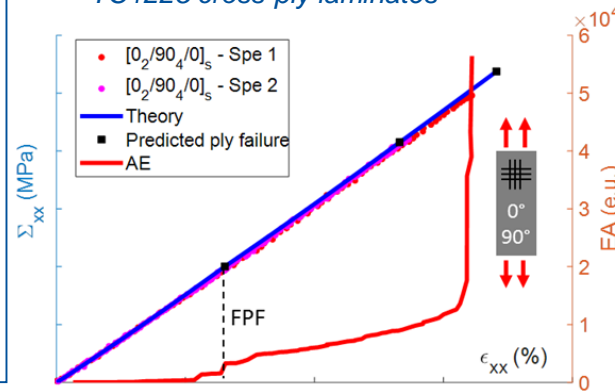
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Identification and validation

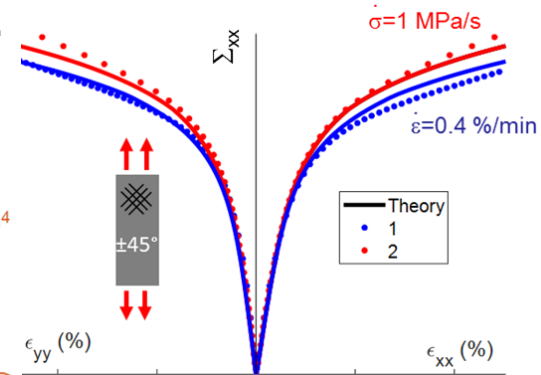
- Already identified and validated on a Carbon/Thermoplastic **TC1225** [Laurin 21]



Macroscopic behaviour of TC1225 cross-ply laminates



Tensile test on TC1225 [0]_{8s}



Macroscopic behaviour of TC1225 [±45]_{2s}

Micromechanical approach

Estimation of longitudinal compressive strength of UD ply

Microscale modelling

Analytical formula

Based on FE simulations at microscale

[Drapier 99], [Gardin 02], [Grandidier 12], [Mechin 18]

- Two main aspects are considered in the proposed model

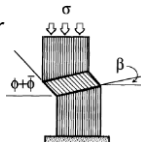
$$\sigma_{UD}^{crit} = \frac{1}{1 + n_{UD} \left(\frac{3}{7}\right)^{\frac{1}{n_{UD}}} \left(\frac{\phi_0/\gamma_{UD}^y}{n_{UD} - 1}\right)^{\frac{n_{UD}-1}{n_{UD}}}} + 2r_{gf} \frac{\pi}{e_b} \sqrt{\frac{E_m E_f}{1 - \nu_m^2} V_f (1 - V_f)}$$

Micro-buckling mechanism

- Fibre buckling [Budiansky 93]
- Plasticity of matrix

Ramberg Osgood behaviour

$$\frac{\gamma}{\gamma_{UD}^y} = \frac{\tau}{\tau_{UD}^y} + \frac{3}{7} \left(\frac{\tau}{\tau_{UD}^y}\right)^{n_{UD}}$$

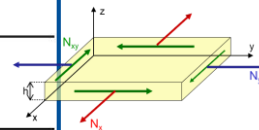


Structural effect

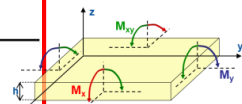
- Ply thickness
- Inner & outer plies
- Different loadings
- Compression & Bending

Ply location	Load	Instability model characteristic thickness : e_b
At the middle of the stacking	Pure compression	$e_b = e$
At one edge of the stacking	Pure compression	$e_b = 2e$
Unidirectional composite	Pure compression	$e_b = \infty$
At the middle of the stacking or at the compressed edge	Bending and compression Neutral plane	$e_b = e$
At the middle of the stacking	Bending with neutral plane Neutral plane	$e_b = e_c$
At one edge of the stacking	Bending Neutral plane	$e_b = e$
Unidirectional composite	Bending	$e_b = c_t e_c$ $c_b \approx 0.4$

Compression



Compression + Bending



Bending

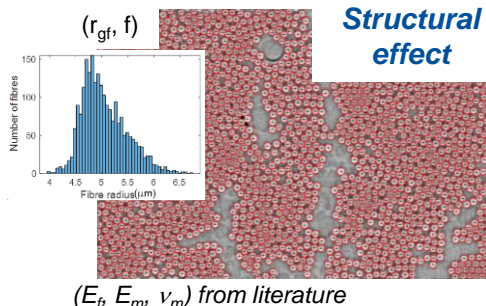
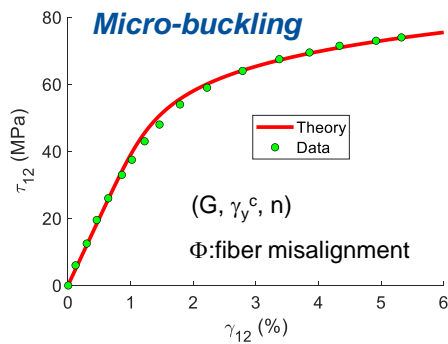
[Grandidier 12]

Analysis of available tests on T700GC/M21

Compression vs. Bending loading

Comparison with experimental data

Identification of micromodel on a Carbon/Epoxy

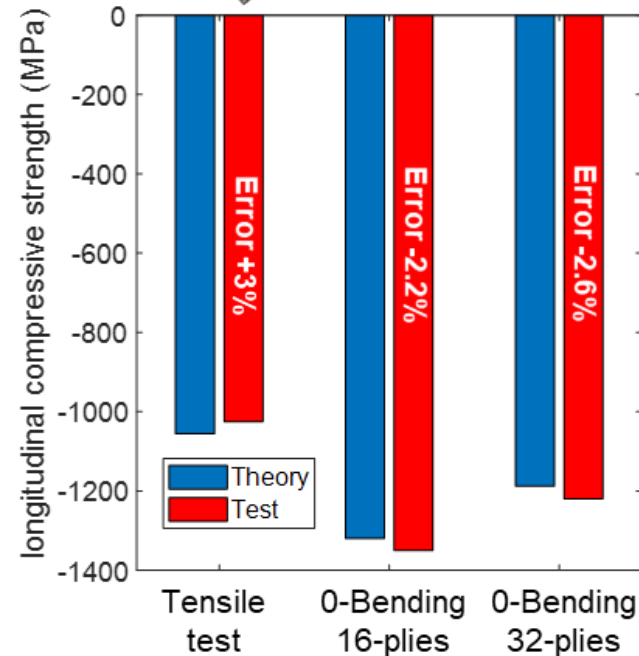


Validation on bending and compression tests

- Tensile tests on laminate which failed in compression (*reference*)
- Bending loadings applied on UD plies [Laurin 16, Laurin 18]

Micromodel analysis agrees with test data

Reference — +30% — +20%

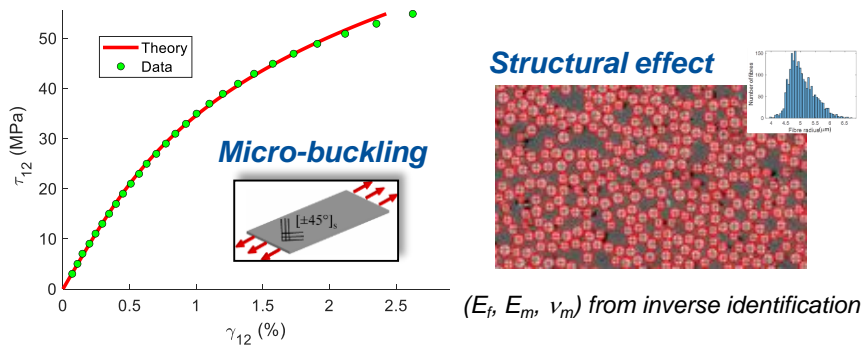


Analysis of available tests on TC1225

Influence of 0-ply position through the thickness

Comparison with experimental data

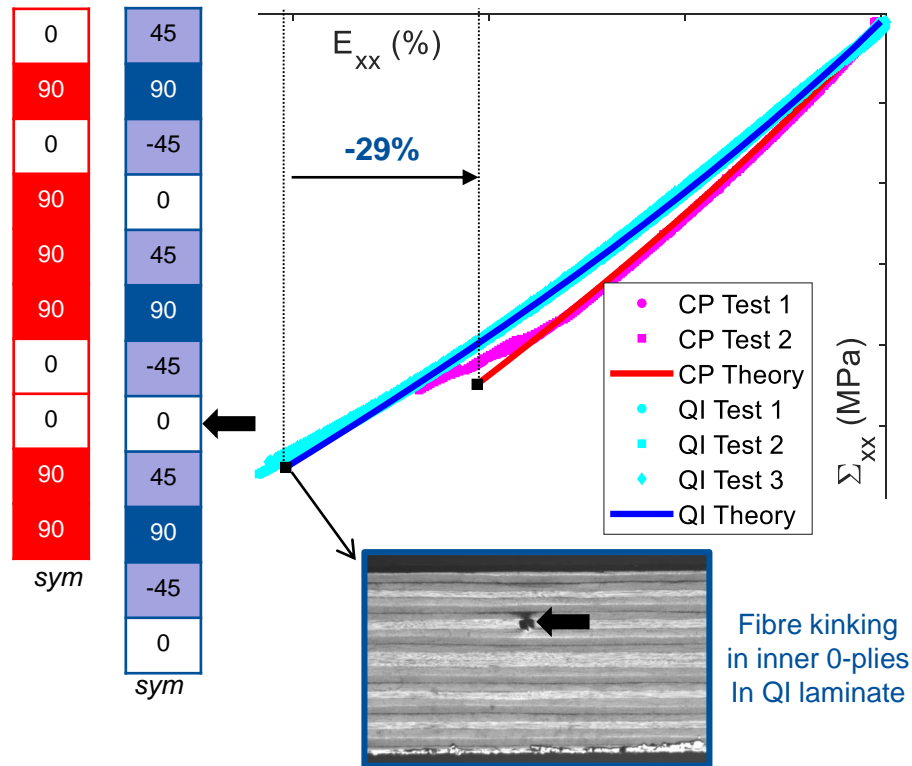
Identification on Carbon/Thermoplastic



Validation different compression tests

- **Cross-Ply** and **Quasi-Isotropic** laminates [Fougerouse 23]
- Different strains at failure for inner and outer 0-ply

Micromodel analysis agrees with test data

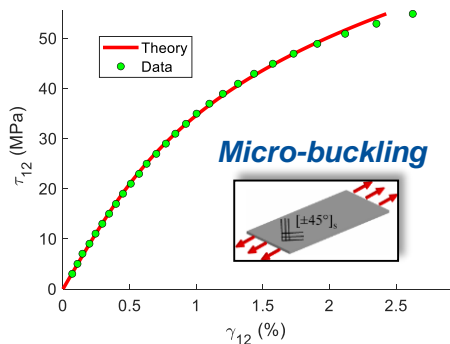


Analysis of available tests on TC1225

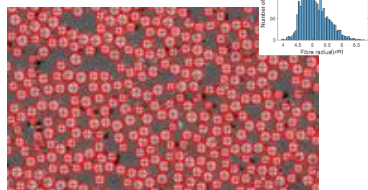
Influence of 0-ply position through the thickness

Comparison with experimental data

Identification on Carbon/Thermoplastic



Structural effect



(E_f, E_m, ν_m) from inverse identification

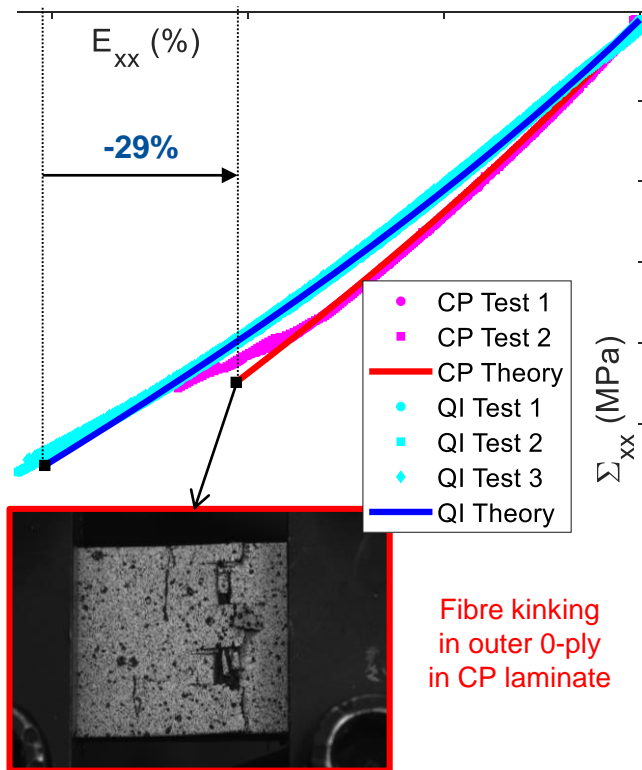
Validation different compression tests

- **Cross-Ply** and **Quasi-Isotropic** laminates [Fougerouse 23]
- Different strains at failure for inner and outer 0-ply

→ Perspectives: [Laurin 18], [Wu 23]

- To consider different bending loadings on TC1225

0	45
90	90
0	-45
90	0
90	45
90	90
0	-45
0	0
90	45
90	90
sym	-45
	sym
	0



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